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In recent years, with the rapid development of small satellites, low-power arcjet thrusters of 100W-class have attracted more attention from the researchers, being simple in structure, compatible with the satellite system, and so on. Ammonia has the advantages of high density, ease of liquefaction and carrying on board, and relatively high specific impulse, thus could be considered as a kind of suitable propellant for propulsion. There have been studies on the ammonia arcjet thrusters at electric power around kilowatt-class or more than 20 kilowatts in the past^[1-3], but studies at electric power below 100W are still scarce. It is not easy to achieve stable discharge state under small gas flow rate and small current conditions with ammonia propellant, and work needs to be done in order to obtain the stable discharge state under such working conditions.

In this study, a low-power arcjet thruster of 100W-class with natural-radiation-cooled nozzle was fired in a vacuum chamber at pressure kept below 1 Pa, and the thrust produced by the arcjet thruster was indirectly measured by the impinging force method^[4]. Thruster performance, i.e., specific impulse and thrust efficiency, and discharge characteristics of the arcjet thruster have been systematically studied. The photos of the arc discharge within the nozzle throat were taken by using a 45° inclined copper mirror and an ICCD camera. Experimental results show that the arcjet thruster can be stably operated with good repeatability. The maximum specific impulse of the thruster is up to 340s, and the maximum thrust efficiency exceeds 40%. The end-on photos indicate that the arc column passes through the nozzle throat and the arc-root is circumferentially attached to the expansion section of the nozzle. Compared to cold NH₃ thruster, the arc-heated one may hold certain advantages in prospective applications on small satellites.

Keywords: Low-power arcjet thruster, Ammonia propellant, Specific impulse, Thrust efficiency, Discharge characteristics

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